

Abstract

A combustion chamber (4) of a gas turbine (1), the combustion space (24) of which is bounded by an annular combustion chamber inner wall (28) and a combustion chamber outer wall (26), in which in order to generate a working medium (M) a supplied fuel is brought into reaction with supplied combustion air, and the combustion chamber wall (25) of which is provided on the inside with a lining formed from a plurality of heat shield elements (38), with the or each heat shield element (38) together with the combustion chamber wall (25) forming an inner space (40) to which a cooling medium (K) can be applied, is to be designed so as to provide a high level of system efficiency at the same time as having a comparatively simple structure and it should also be possible to disassemble the combustion chamber inner wall (28) in a time-saving manner. For this purpose according to the invention there is disposed in each case in the respective inner space (40) a cooling medium distributor (42) via which a cooling medium supply line (44) is connected to a plurality of cooling medium exit openings (46) and the combustion chamber inner wall (28) is formed from a plurality of wall elements (30) abutting each other at a horizontal parting joint (31), whereby the abutting wall elements (30) of the combustion chamber inner wall (28) are connected to one another at their horizontal parting joint (31) by means of a plurality of screw connections (32) oriented at an angle to the inner wall surface.

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FIG 1